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Adams et al.(10) **Pub. No.: US 2012/0291449 A1**(43) **Pub. Date: Nov. 22, 2012**(54) **TURBINE SECTION OF HIGH BYPASS
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17, 2011, provisional application No. 61/593,190,
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F02C 3/04 (2006.01)(52) **U.S. Cl.** **60/793**(57) **ABSTRACT**

A turbofan engine has an engine case and a gaspath through the engine case. A fan has a circumferential array of fan blades. The engine further has a compressor, a combustor, a gas generating turbine, and a low pressure turbine section. A speed reduction mechanism couples the low pressure turbine section to the fan. A bypass area ratio is greater than about 6.0. The low pressure turbine section airfoil count to bypass area ratio is below about 170.

